

SUPPLEMENTAL TYPE CERTIFICATE

10052286 REV. 4

This Supplemental Type Certificate is issued by EASA, acting in accordance with Regulation (EC) No. 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation and in accordance with Commission Regulation (EU) No. 748/2012 to:

BERINGER AERO S.A.S.

**AEROPOLE
05130 TALLARD
FRANCE**

and certifies that the change in the type design for the product listed below with the limitations and conditions specified meets the applicable Type Certification Basis and environmental protection requirements when operated within the conditions and limitations specified below:

Original Type Certificate Number: EASA.A.362

Type Certificate Holder: EXTRA Flugzeugproduktions-

Type: EA 300

Model: EA 300, EA 300/200

EA 300/L, EA 300/LC

EA 300/LT, EA 300/S

EA 300/SC

Description of Design Change:

Replacement of wheel and brake system

Rev.1 - Update with longer axle bolts

Rev.2 - Update with different hydraulic fitting

Rev.3 - Test Report Improvement

Rev.4 - Update with new wheel part number

See Continuation Sheet(s)

For the European Aviation Safety Agency

Date of Issue: 30 January 2017



Dominique ROLAND
Head of General Aviation and
Remotely Piloted Aircraft Systems (RPAS)

10035860

SUPPLEMENTAL TYPE CERTIFICATE - 10052286 - REV. 4 - BERINGER AERO S.A.S. - 303409

EASA Certification Basis:

The Certification Basis (CB) for the original product remains applicable to this certificate/ approval.

The requirements for environmental protection and the associated certified noise and/ or emissions levels of the original product are unchanged and remain applicable to this certificate/ approval.

Associated Technical Documentation:

- Dossier de modification DM-STC-12 Rev. 5 dated 3 Jan 2017
- Maintenance and overhaul manual MM-STC-012 Rev. 3 dated 18 Jan 2017 or later revisions of the above listed documents approved by EASA.

Limitations/Conditions:

The approval holder shall fulfill the obligations of Part 21, Point 21.A.109 or 21.A.451(b) (as applicable).

Prior to installation of this design change it must be determined that the interrelationship between this design change and any other previously installed design change and/ or repair will introduce no adverse effect upon the airworthiness of the product.

- End -